

A BENCHMARK PROBLEM FOR NONLINEAR CONTROL DESIGN

ROBERT T. BUPP^{1†}, DENNIS S. BERNSTEIN^{2*†} AND VINCENT T. COPPOLA^{2‡}

¹*TRW, One Space Park, Mail Stop R9/1944H, Redondo Beach, CA 90278, U.S.A.*

²*Department of Aerospace Engineering, The University of Michigan, Ann Arbor, MI, U.S.A*

1. INTRODUCTION

This paper describes a nonlinear control design problem involving the nonlinear interaction of a translational oscillator and an eccentric rotational proof mass. This problem provides a benchmark for examining nonlinear control design techniques within the framework of a nonlinear fourth-order dynamical system. The problem is posed in the spirit of the linear benchmark problem described in Reference 1.

This system was originally studied as a simplified model of a dual-spin spacecraft to investigate the resonance capture phenomenon.² More recently, it has been studied to investigate the utility of a rotational proof-mass actuator for stabilizing translational motion.^{3–5} Viewed in this way, the rotational/translational proof-mass actuator (RTAC) has the feature that the nonlinearities associated with the actuator stroke limitation are implicit in the system dynamics. In contrast, the stroke limitation constraint must be considered separately in linear translational proof-mass actuators.⁶ A similar system has been studied as a rotating unbalanced mass (RUM) actuator in References 7 and 8.

2. PROBLEM STATEMENT

The system shown in Figure 1 represents a translational oscillator with an eccentric rotational proof-mass actuator. The oscillator consists of a cart of mass M connected to a fixed wall by a linear spring of stiffness k . The cart is constrained to have one-dimensional travel. The proof-mass actuator attached to the cart has mass m and moment of inertia I about its centre of mass, which is located a distance e from the point about which the proof mass rotates. The motion occurs in a horizontal plane, so that no gravitational forces need to be considered. In Figure 1, N denotes the control torque applied to the proof mass, and F is the disturbance force on the cart.

Let q and \dot{q} denote the translational position and velocity of the cart, and let θ and $\dot{\theta}$ denote the angular position and velocity of the rotational proof mass, where $\theta = 0$ is perpendicular to the

* Correspondence to: Dennis S. Bernstein, Department of Aerospace Engineering, University of Michigan, 1320 Beal Street, Ann Arbor, MI 48109-2140, U.S.A. E-mail: dsbaero@engin.umich.edu

† Research supported in part by the Air Force Office of Scientific Research under Grant F49620-95-1-0019.

‡ Research supported in part by NSF Grant MSS-9309165.

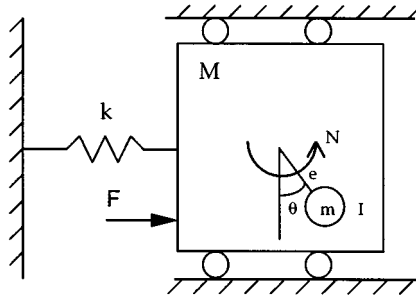


Figure 1. Rotational actuator to control a translational oscillator

motion of the cart, and $\theta = 90^\circ$ is aligned with the positive q direction. The equations of motion are given by

$$(M + m)\ddot{q} + kq = -me(\ddot{\theta} \cos \theta - \dot{\theta}^2 \sin \theta) + F$$

$$(I + me^2)\ddot{\theta} = -me\ddot{q} \cos \theta + N$$

With the normalizations³

$$\xi \triangleq \sqrt{\frac{M + m}{I + me^2}} q, \quad \tau \triangleq \sqrt{\frac{k}{M + m}} t$$

$$u \triangleq \frac{M + m}{k(I + me^2)} N, \quad w \triangleq \frac{1}{k} \sqrt{\frac{M + m}{I + me^2}} F,$$

the equations of motion become

$$\ddot{\xi} + \xi = \varepsilon \left(\dot{\theta}^2 \sin \theta - \ddot{\theta} \cos \theta \right) + w$$

$$\ddot{\theta} = -\varepsilon \ddot{\xi} \cos \theta + u$$

where ξ is the normalized cart position, and w and u represent the non-dimensionalized disturbance and control torque, respectively. In the normalized equations, the symbol (\cdot) represents differentiation with respect to the normalized time τ . The coupling between the translational and rotational motions is represented by the parameter ε which is defined by

$$\varepsilon \triangleq \frac{me}{\sqrt{(I + me^2)(M + m)}}$$

Letting $x = [x_1, x_2, x_3, x_4]^T = [\xi, \dot{\xi}, \theta, \dot{\theta}]^T$, the non-dimensional equations of motion in first-order form are given by

$$\dot{x} = f(x) + g(x) u + d(x) w,$$

where

$$f(x) = \begin{bmatrix} x_2 \\ \frac{-x_1 + \varepsilon x_4^2 \sin x_3}{1 - \varepsilon^2 \cos^2 x_3} \\ x_4 \\ \frac{\varepsilon \cos x_3 (x_1 - \varepsilon x_4^2 \sin x_3)}{1 - \varepsilon^2 \cos^2 x_3} \end{bmatrix}, \quad g(x) = \begin{bmatrix} 0 \\ \frac{-\varepsilon \cos x_3}{1 - \varepsilon^2 \cos^2 x_3} \\ 0 \\ \frac{1}{1 - \varepsilon^2 \cos^2 x_3} \end{bmatrix}, \quad d(x) = \begin{bmatrix} 0 \\ \frac{1}{1 - \varepsilon^2 \cos^2 x_3} \\ 0 \\ \frac{-\varepsilon \cos x_3}{1 - \varepsilon^2 \cos^2 x_3} \end{bmatrix}$$

The nonlinear benchmark problem can now be stated as follows.

Nonlinear benchmark problem

Design a controller that satisfies the following criteria:

1. The closed-loop system is stable (e.g. locally or globally).
2. The closed-loop system exhibits good settling behaviour for a class of initial conditions.
3. The closed-loop system exhibits good disturbance rejection compared to the uncontrolled oscillator for a class of disturbance signals.
4. The control effort should be reasonable (e.g. maximum torque).
It may be interesting to consider the following, optional objective:
5. The controller should not distinguish between the values θ and $\theta \bmod 2\pi$, since these values represent the same rotational configuration.

The requirements 1–4 for stabilization, free response, disturbance rejection and control effort are not precisely stated. Instead, each designer is given some freedom to interpret these issues individually. Requirement 5 avoids ‘unwinding’, i.e. the use of control effort to move the system from $[0, 0, 2\pi n, 0]^T$ to $[0, 0, 0, 0]^T$. Additionally, each designer may impose additional constraints on the problem as desired, where such features serve to highlight the capabilities of particular nonlinear control design methods.

3. LABORATORY TESTBED

A laboratory-scale version of the nonlinear benchmark problem has been constructed.⁹ The parameters for a nominal configuration are given in Table I.

Table I

Description	Parameter	Value	Units
Cart mass	M	1.3608	kg
Arm mass	m	0.096	kg
Arm eccentricity	e	0.0592	m
Arm inertia	I	0.0002175	kg m ²
Spring stiffness	k	186.3	N/m
Coupling parameter	ε	0.200	—

The physical configuration of the system necessitates the constraint

$$|q| \leq 0.025 \text{ m}$$

In addition, the control torque is limited by $N \leq 0.100 \text{ N m}$ continuous, although somewhat higher torques can be tolerated for short periods.

REFERENCES

1. Wie, B. and D. S. Bernstein 'Benchmark problems in robust control design', *J. Guidance Control Dyn.*, **15**, 1057–1059 (1992).
2. Rand, R. H., R. J. Kinsey and D. L. Mingori, 'Dynamics of spinup through resonance', *Int. J. Non-Linear Mech.*, **27**, 489–502 (1992).
3. Wan, C. J., D. S. Bernstein and V. T. Coppola, 'Global stabilization of the oscillating eccentric rotor', *Proc. IEEE Conf. Decision and Control*, Orlando, FL, 1994, pp. 4024–4029. Also, *Nonlinear Dyn.*, **10**, 49–62 (1996).
4. Bupp, R. T., C. J. Wan, V. T. Coppola and D. S. Bernstein, 'Design of a rotational actuator for global stabilization of translational motion', *Proc. ASME Winter Meeting*, DE-Vol. 75, Chicago, IL, 1994, pp. 449–456.
5. Bupp, R. T., V. T. Coppola and D. S. Bernstein, 'Vibration suppression of multi-modal translational motion using a rotational actuator', *Proc. IEEE Conf. Decision and Control*, Orlando, FL, 1994, pp. 4030–4034.
6. Lindner, D. K., T. P. Celano and E. N. Ide, 'Vibration suppression using a proofmass actuator operating in stroke/force saturation', *J. Vib. Acoust.*, **113**, 423–433 (1991).
7. Polites, M. E., Rotating-Unbalanced-Mass devices for scanning balloon-borne experiments, free-flying spacecraft, and space shuttle/space station experiments, Technical Report TP-3030, NASA, NASA Marshall Space Flight Center, AL, 1990.
8. Polites, M. E., 'New method for scanning spacecraft and balloon-borne/space-based experiments', *J. Guidance Control Dyn.*, **14** (3), 548–553 (1991).
9. Bupp, R. T., D. S. Bernstein and V. T. Coppola, 'A benchmark problem for nonlinear control design: problem statement, experimental testbed, and passive nonlinear compensation', *Proc. American Control Conf.*, Seattle, WA, 1995, pp. 4363–4367.